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三维 ALE15 翼型空化流动数值模拟*

谭磊¹ 曹树良² 王玉明¹ 祝宝山²

(1. 清华大学摩擦学国家重点实验室, 北京 100084; 2. 清华大学水沙科学与水利水电工程国家重点实验室, 北京 100084)

【摘要】 在均相流假设下, 考虑流体压力和速度湍流脉动、不可凝结性气体的影响, 采用完全空化模型计算空化流场的相变, 引入密度函数对 RNG $k-\varepsilon$ 湍流模型的湍流粘性系数进行修正, 提出了一种空化流动的数值模型和计算方法。根据试验条件给定的参数, 采用提出的数值模型和计算方法, 数值模拟了空化数为 2.3 时 ALE15 翼型定常空化流动。计算得到的不同剖面速度分布与试验数据吻合较好, 验证了该数值模型和计算方法的一致性。不同剖面上, 远离翼型表面的速度与主流区速度接近, 沿着流动方向, 远离翼型表面的速度逐渐减小, 这与空泡形成的阻碍有关。空泡尾部出现较大的涡流区, 靠近翼型表面的速度为负值, 这与反向射流的作用有关。

关键词: ALE15 翼型 空化模型 湍流模型 数值模拟

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Numerical Simulation of Cavitation for 3-D ALE15 Hydrofoil

Tan Lei¹ Cao Shuliang² Wang Yuming¹ Zhu Baoshan²

(1. State Key Laboratory of Tribology, Tsinghua University, Beijing 100084, China

2. State Key Laboratory of Hydrosience and Engineering, Tsinghua University, Beijing 100084, China)

Abstract

The phase transformation in cavitation flow field was calculated by the full cavitation model considering the pressure and velocity turbulent fluctuation of the fluid, as well as the influence of noncondensable gas based on the homogeneous flow assumption. The turbulence viscosity coefficient was modified by the density function. A computation model and calculation method was proposed for the steady cavitation. The steady cavitation flow field of ALE15 hydrofoil was numerical simulated for the cavitation numbers of 2.3 by using the computation model and calculation method according to the conditions in the experiment. The calculated velocity distributions on different profiles agreed well with the experiment data, which validated the reliability of this computation model and calculation method. The velocity far away from the hydrofoil section was close to the velocity in the main flow region, and gradually decreased along the flow direction for the reason that the cavity acted as an obstacle. There is a large vortex zone in the cavity rear, and the velocity near the hydrofoil section is negative, both caused by the re-entrant jet.

Key words ALE15 hydrofoil, Cavitation model, Turbulence model, Numerical simulation

引言

空化是涉及相变的可压缩、非非常的复杂流动。为深入研究空化流动的机理, 许多学者采用试验测

量和数值模拟开展了一系列工作^[1-11]。在均相流假设下, 根据计算流体介质密度方法的不同, 可将空化模型分为空泡动力学模型、状态方程模型和输运方程模型 3 类。Kubota^[12]、Merkle^[13]、Kunz^[14]、

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作者简介: 谭磊, 博士后, 主要从事流体机械设计研究, E-mail: tanlei@mail.tsinghua.edu.cn

通讯作者: 曹树良, 教授, 博士生导师, 主要从事流体机械设计研究, E-mail: caoshl@mail.tsinghua.edu.cn

Singhal^[15]等给出了输运方程中源项的不同表达。湍流模型对空化数值计算中空泡的生成和溃灭有很大影响,Coutier^[16]和 Johansen^[17]给出了空化流动中湍流模型的不同修正。

本文采用完全空化模型和密度修正的 RNG $k-\varepsilon$ 湍流模型数值模拟 ALE15 翼型的空化流动,与试验数据对比分析数值模拟的结果。

1 数学模型及计算方法

1.1 流体的连续方程及运动方程

在均相流假设下,流场内的流体可视为均匀的可压缩气液混合相介质,气液混合相介质在流场中任一点的速度和压力相同。气液混合相介质的连续方程和运动方程为

$$\frac{\partial}{\partial t}(\rho_m) + \nabla \cdot (\rho_m \mathbf{u}) = 0 \quad (1)$$

$$\frac{\partial}{\partial t}(\rho_m \mathbf{u}) + \nabla \cdot (\rho_m \mathbf{u} \mathbf{u}) = -\nabla p + \nabla \cdot [(\mu_m + \mu_t) \nabla \mathbf{u}] + \frac{1}{3} \nabla [(\mu_m + \mu_t) \nabla \cdot \mathbf{u}] \quad (2)$$

式中 ρ_m ——混合相的密度

p ——压力 \mathbf{u} ——速度矢量

μ_t ——涡旋粘性系数

μ_m ——混合相的动力粘性系数,按气、液两相体积分数加权平均后得到

1.2 空化模型方程

采用 Singhal^[15]提出的完全空化模型,混合相的密度 ρ_m 定义为

$$\frac{1}{\rho_m} = \frac{f_v}{\rho_v} + \frac{f_g}{\rho_g} + \frac{f_l}{\rho_l} \quad (3)$$

$$\begin{cases} f_v = \frac{\alpha_v \rho_v}{\rho_m} \\ f_g = \frac{\alpha_g \rho_g}{\rho_m} \\ f_l = \frac{\alpha_l \rho_l}{\rho_m} = 1 - f_v - f_g \end{cases} \quad (4)$$

式中 ρ_v, ρ_g, ρ_l ——汽相、不可凝结气相、液相密度

f_v, f_g, f_l ——各相对应的质量分数

$\alpha_v, \alpha_g, \alpha_l$ ——各相对应的体积分数

气相组分输运方程为

$$\begin{aligned} \frac{\partial}{\partial t}(\alpha_v \rho_v) + \nabla \cdot (\alpha_v \rho_v \mathbf{u}) = \\ C_c \frac{\sqrt{k}}{\lambda} \rho_l \rho_v \left(\frac{2p_v - p}{3\rho_l} \right)^{\frac{1}{2}} (1 - f_v - f_g) - \\ C_c \frac{\sqrt{k}}{\lambda} \rho_l \rho_l \left(\frac{2p - p_v}{3\rho} \right)^{\frac{1}{2}} f_v \end{aligned} \quad (5)$$

其中 $f_g = 1.5 \times 10^{-5}$

式中 C_e, C_c ——经验常数,分别取 0.02、0.01

p_v ——汽化压强

k ——湍动能

λ ——液相表面张力系数,取 0.071 7 N/m

1.3 湍流模型方程

采用 RNG $k-\varepsilon$ 双方程湍流模型,其中 k 和 ε 方程为

$$\frac{\partial(\rho_m k)}{\partial t} + \frac{\partial(\rho_m u_i k)}{\partial x_i} = G_k + \rho_m \varepsilon + \frac{\partial}{\partial x_j} \left[\alpha_k (\mu_m + \mu_t) \frac{\partial k}{\partial x_j} \right] \quad (6)$$

$$\frac{\partial(\rho_m \varepsilon)}{\partial t} + \frac{\partial(\rho_m u_i \varepsilon)}{\partial x_i} = C_{1\varepsilon} \frac{\varepsilon}{k} G_k - C_{2\varepsilon} \rho_m \frac{\varepsilon^2}{k} + \frac{\partial}{\partial x_j} \left[\alpha_\varepsilon (\mu_m + \mu_t) \frac{\partial \varepsilon}{\partial x_j} \right] \quad (7)$$

$$\mu_t = \rho_m c_\mu \frac{k^2}{\varepsilon} \quad (8)$$

其中 $\alpha_k = \alpha_\varepsilon = 1.39$ $c_\mu = 0.09$

式中 ε ——湍动能耗散率

G_k ——湍动能生成项

$C_{1\varepsilon}, C_{2\varepsilon}$ ——经验常数

u_i ——速度分量

x_i, x_j ——张量形式坐标分量

$\alpha_k, \alpha_\varepsilon, c_\mu$ ——经验常数,分别取 1.39、1.39、0.09

考虑空泡流可压缩性的影响,引入函数 $f(\rho_m)$ 对 μ_t 进行修正

$$\mu_t = f(\rho_m) c_\mu \frac{k^2}{\varepsilon} \quad (9)$$

$f(\rho_m)$ 定义为

$$f(\rho_m) = \rho_v + [(\rho_m - \rho_v) / (\rho_l - \rho_v)]^n (\rho_l - \rho_v)$$

其中 n 为常数,当 n 取不同值时, $f(\rho_m)$ 与混合物密度 ρ_m 之间的关系如图 1 所示,增大 n 的取值可有效减少混合物的湍流粘性系数,本文计算中取 $n = 10$ 。

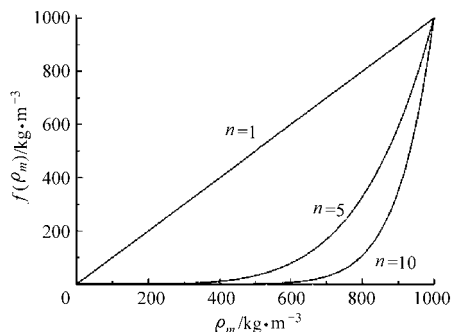


图1 修正函数与混合物密度的关系曲线
Fig.1 Relationship curves of modified function and mixture density

1.4 计算区域及网格

采用文献[1]中的 ALE15 翼型进行数值模拟, 其几何结构如下: 翼型弦长 107.9 mm, 宽度(z 方向)和厚度(y 方向)分别为 50 mm 和 16 mm, 翼型进口边与 z 轴夹角为 15° 。采用结构化六面体网格对计算域进行网格划分, 翼型头、尾部距计算域进、出口分别为 10 倍弦长, 如图 2 所示。

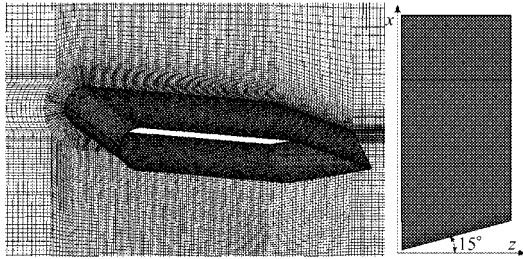


图2 三维 ALE15 翼型计算域及网格

Fig.2 Computational domain and grid of ALE15

1.5 数值方法及参数设置

雷诺(Reynolds)数 Re 和空化数 σ 分别定义为

$$Re = \rho_l v_\infty C / \mu_l \quad (10)$$

$$\sigma = (p_\infty - p_v) / (0.5 \rho_l v_\infty^2) \quad (11)$$

式中 v_∞ —— 来流速度

μ_l —— 液相动力粘性系数

p_∞ —— 计算域参考压力

按文献[1]给定流体的物性参数和计算工况: 液相和汽相密度分别为 998.2 kg/m^3 和 0.554 kg/m^3 , 翼型攻角为 5° , 进口速度为 13 m/s , 空化数 σ 为 2.3, 雷诺数 Re 为 1.4×10^6 。

计算域进、出口及壁面的边界条件: 进口给定均

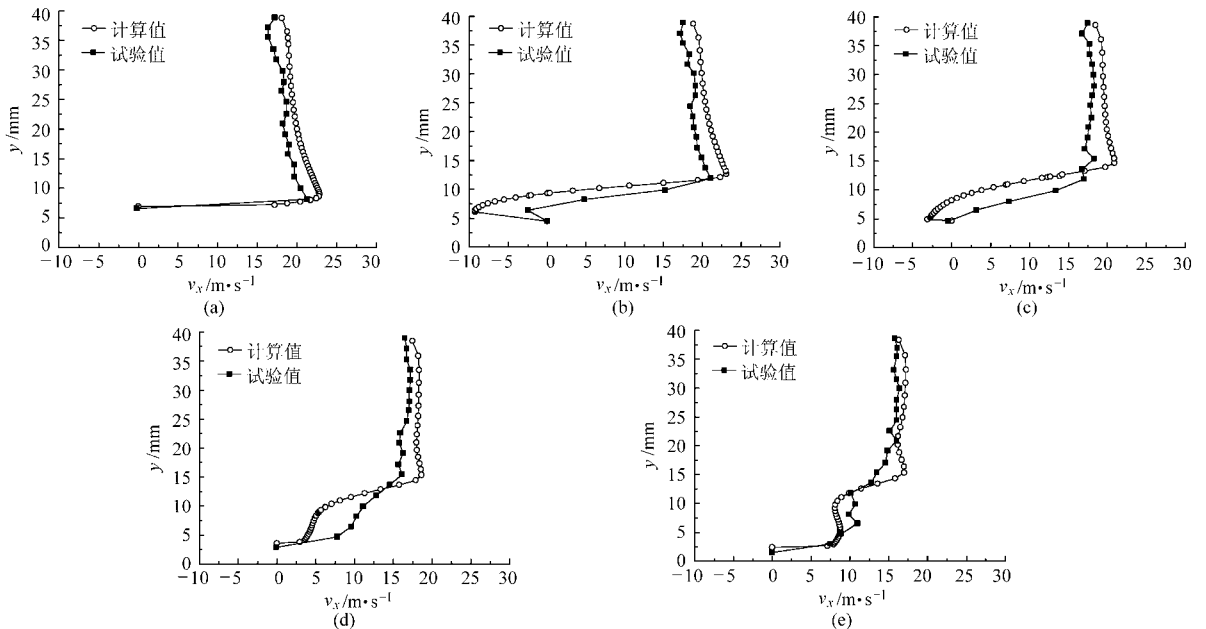


图5 不同剖面的速度分布曲线

Fig.5 Velocity distribution on profiles

(a) $x = 0 \text{ mm}$ (b) $x = 13 \text{ mm}$ (c) $x = 26 \text{ mm}$ (d) $x = 39 \text{ mm}$ (e) $x = 52 \text{ mm}$

匀来流速度, 出口给定压力, 其他变量给定充分发展条件, 对流项采用二阶迎风格式离散, 其他项采用中心差分格式离散。

2 计算结果及分析

为分析数值模拟结果, 在 $z = 45 \text{ mm}$ 的截面上, 布置如图 3 所示的计算剖面。

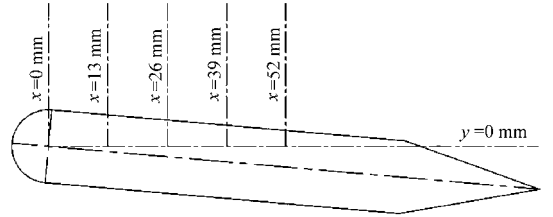


图3 三维翼型 ALE15 计算剖面图

Fig.3 Computational profile of 3-D ALE15 hydrofoil

2.1 空泡形状

图 4 给出了数值模拟得到的翼型吸力面空泡形状, 沿 z 轴方向空泡形状不同, 表现出较强的三维特征, 空泡尾部在靠近翼型吸力面附近出现了回缩。

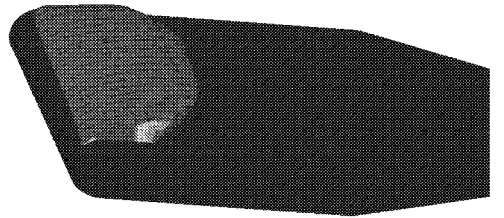


图4 翼型表面空泡形状

Fig.4 Cavity shape on hydrofoil section

2.2 不同剖面的速度分布

图 5 给出了 $z = 45 \text{ mm}$ 截面上, 5 个不同剖面

$x = 0 \text{ mm}$ 、 $x = 13 \text{ mm}$ 、 $x = 26 \text{ mm}$ 、 $x = 39 \text{ mm}$ 和 $x = 52 \text{ mm}$ 上的速度分布。由图可知,采用本文数值模型和计算方法得到的计算结果与试验数据吻合较好。

在 $x = 0 \text{ mm}$ 剖面上,空泡内部靠近翼型吸力面的速度接近零,由于空泡的头部很薄,随着 y 值增加,速度很快增大为主流区速度。在 $x = 13 \text{ mm}$ 剖面上,靠近翼型吸力面的速度为负值,原因是此剖面接近空泡尾部,空泡尾部出现反向射流,导致速度为负值。沿着流动方向,随着 x 值的增大, $x = 26 \text{ mm}$ 、 $x = 39 \text{ mm}$ 和 $x = 52 \text{ mm}$ 剖面上远离翼型表面的主流区速度逐渐减小,可能的原因是翼型吸力面上的空泡对流动形成了阻碍,导致速度减小。

2.3 翼型表面的速度分布

图6给出了 $z = 45 \text{ mm}$ 截面上翼型表面的速度分布,等值线的数值为混合介质密度。由图可知,在空泡壁面上部,流线分布非常均匀,这与图5中不同剖面上 y 值较大时,流体速度与主流区速度相等是一致的。在空泡尾部出现了较大漩涡,靠近翼型吸力面的速度为负值,这是由于反向射流的作用。

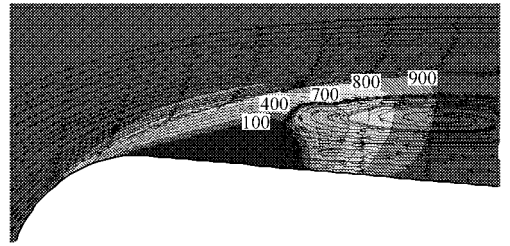


图6 $z = 45 \text{ mm}$ 截面速度分布

Fig. 6 Velocity distribution around foil on $z = 45 \text{ mm}$

3 结论

(1)在均相流假设下,采用完全空化模型计算空化流场的相变,采用密度函数对 RNG $k - \varepsilon$ 湍流模型中的湍流粘性系数进行了修正,发展了一种空化流场的数值模拟方法。

(2)数值模拟了三维 ALE15 翼型的定常空化流动,计算得到的空泡形状具有明显的三维特征,不同剖面上的速度与试验数据吻合较好,验证了本文提出的数值模型和计算方法的准确性。分析了翼型表面的速度分布,发现空泡尾部的反向射流是引起空泡尾部漩涡的原因。

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